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REPORT NO. AE61-1055

DATE 8 Feb. 1961

NO. OF PAGES 18

CONVAIR ASTRONAUTICS

CONVAIR DIVISION OF GENERAL DYNAMICS CORPORATION

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GENERAL FLIGHT TEST PLAN

SCIENTIFIC PASSENGER POD PROGRAM

ATLANTIC MISSILE RANGE

AE61-1055, SUPPLEMENT 1

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FOREWORD

In order to maintain the classification of the general test plan and all individual pod flight test plans for the Scientific Passenger Pod Program at a confidential level, Section 3.3 (Missile Performance) of the general flight test plan has been extracted from the report and the information presented in the following supplement.

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### 3.3 MISSILE PERFORMANCE

Approximate flight performance parameters for Series E and F R&D missiles carrying the scientific passenger pod are given in Figures 1, 2, 3, 4, 5, 6, and 7. The parameters presented in these figures and in Table I were calculated on the basis of the following assumptions:

1. Launch azimuth of  $106.3^{\circ}$  true from Complex 11 or 13.
2. Impact at 4400 NM nominal range at  $8.3^{\circ}$  S latitude and  $14.4^{\circ}$  W longitude.
- 2a. Impact at 6350 NM nominal range at  $28.0^{\circ}$  S latitude and  $13.2^{\circ}$  E longitude. (Applicable to Fig. 7 only.)
3. Rotating earth.
4. NPSH, dynamic pressure, aerodynamic heating, and maximum sustainer duration constraints satisfied.
5. Nominal MA-3 propulsion system performance.
6. Current weights of Series E R&D missile.
7. Series E tilt program with attenuation factor of 0.97.
8. Free launch.
9. The seven different configurations with their corresponding weights are listed at the end of Table I.
10. Vacuum re-entry.
11. Lock-on angle of  $20^{\circ}$ .
- 11a. Lock-on angle of  $28.3^{\circ}$ . (Applicable to Fig. 6 only.)
- 11b. Lock on angle of  $17^{\circ}$ . (Applicable to Fig. 7 only.)

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TABLE I      SERIES E R&D (WITH SP POD)

PARAMETER	a	
<u>LAUNCH</u>		
Weight	267,308 lbs.	26'
Booster engine thrust (sea level)	334,415 lbs.	33'
Sustainer engine thrust (sea level)	56,340 lbs	50'
Vernier engine thrust (sea level)	1,983 lbs	
Axial lift-off thrust (sea level)	392,738 lbs	39'
Initial thrust-to-weight ratio	1.47 g's	
Booster propellant flow at lift-off	1318.76 lb/sec	131'
Sustainer propellant flow at lift-off	263.16 lb/sec	26'
Vernier propellant flow at lift-off	9.64 lb/sec	
<u>BOOSTER CUTOFF</u>		
Time	123.58 sec	12'
Weight after jettisoning booster section	60,677 lbs	60'
Velocity (relative to earth)	8,938 ft/sec	8'
Altitude	205,098 ft	205'
Flight path angle (relative)	29.1°	
Acceleration	7.02 g's	7'
Range	41.3 NM	41'
Sustainer propellant flow (after separation)	254.87 lb/sec	25'
Vernier propellant flow (after separation)	9.49 lb/sec	
<u>SUSTAINER CUTOFF</u>		
Time	297.09 sec	29'
Weight	15,525 lbs	15'
Velocity (relative to earth)	20,466 ft/sec	20'
Altitude	157.7 NM	15'
Flight path angle (relative)	17.8°	1'
Acceleration	5.16 g's	
Range	385.4 NM	38'
<u>VERNIER CUTOFF</u>		
Time	313.06 sec	31'
Weight	15,399 lbs	15'
Velocity (relative to earth)	20,367 ft/sec	20'

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WITH SP POD) FLIGHT PARAMETERS

VALUE						
	b	c	d	e	f	g
08 lbs.	267,118 lbs	267,808 lbs	267,708 lbs.	267,893 lbs.	267,708 lbs.	266,518 lbs
15 lbs	334,415 lbs	334,415 lbs	334,415 lbs	334,415 lbs.	334,415 lbs	334,415 lbs
40 lbs	56,340 lbs	56,340 lbs	56,350 lbs	56,340 lbs	56,340 lbs	56,340 lbs
83 lbs	1,983 lbs	1,983 lbs	1,983 lbs	1,983 lbs	1,983 lbs	1,983 lbs
38 lbs	392,738 lbs	392,738 lbs	392,738 lbs	392,738 lbs	392,738 lbs	392,738 lbs
47 g's	1.47 g's	1.47 g's	1.47 g's	1.47 g's	1.47 g's	1.47 g's
76 lb/sec	1318.76 lb/sec	1318.76 lb/sec	1318.76 lb/sec	1318.76 lb/sec	1318.76 lb/sec	1318.76 lb/sec
16 lb/sec	263.16 lb/sec	263.16 lb/sec	263.16 lb/sec	263.16 lb/sec	263.16 lb/sec	263.16 lb/sec
64 lb/sec	9.64 lb/sec	9.64 lb/sec	9.64 lb/sec	9.64 lb/sec	9.64 lb/sec	9.64 lb/sec
5.58 sec	123.49 sec	123.84 sec	123.79 sec	123.87 sec	123.79 sec	123.19 sec
377 lbs	60,628 lbs	60,793 lbs	60,768 lbs	60,826 lbs	60,768 lbs	60,475 lbs
38 ft/sec	8,939 ft/sec	8,936 ft/sec	8,937 ft/sec	8,934 ft/sec	8,937 ft/sec	8,943 ft/sec
98 ft	205,162 ft	205,043 ft	205,061 ft	204,971 ft	205,061 ft	205,206 ft
1°	29.2°	29.1°	29.1°	29.0°	29.1°	29.2°
02 g's	7.02 g's	7.01 g's	7.01 g's	7.00 g's	7.01 g's	7.04 g's
3 NM	41.3 NM	41.4 NM	41.3 NM	41.4 NM	41.3 NM	41.2 NM
87 lb/sec	254.87 lb/sec	254.87 lb/sec	254.87 lb/sec	254.87 lb/sec	254.87 lb/sec	254.87 lb/sec
49 lb/sec	9.49 lb/sec	9.49 lb/sec	9.49 lb/sec	9.49 lb/sec	9.49 lb/sec	9.49 lb/sec
09 sec	296.84 sec	297.70 sec	297.57 sec	297.85 sec	297.95 sec	305.87 sec
525 lbs	15,516 lbs	15,548 lbs	15,543 lbs	15,551 lbs	15,444 lbs	12,892 lbs
466 ft/sec	20,467 ft/sec	20,471 ft/sec	20,470 ft/sec	20,472 ft/sec	20,155 ft/sec	22,266 ft/sec
7 NM	157.8 NM	157.6 NM	157.6 NM	157.6 NM	176.3 NM	160.5 NM
8°	17.8°	17.8°	17.8°	17.8°	23.0°	16.5°
16 g's	5.16 g's	5.16 g's	5.16 g's	5.15 g's	5.19 g's	6.22 g's
4 NM	385 NM	386.4 NM	386.2 NM	386.7 NM	376.7 NM	418.7 NM
06 sec	312.81 sec	313.64 sec	313.51 sec	313.61 sec	313.86 sec	321.84 sec
99 lbs	15,390 lbs	15,422 lbs	15,417 lbs	15,427 lbs	15,318 lbs	12,766 lbs
67 ft/sec	20,367 ft/sec	20,372 ft/sec	20,371 ft/sec	20,374 ft/sec	20,018 ft/sec	22,184 ft/sec

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TABLE I SERIES E R&D (WITH SP POD) FLIGHT

PARAMETER	a	b
<u>VERNIER CUTOFF (Continued)</u>		
Altitude	174 NM	174.1 NM
Flight path angle (relative)	17.5°	17.5°
Acceleration	0.08 g's	0.08 g's
Range	434.2 NM	433.8 NM
<u>APOGEE</u>		
Time	961 sec	961 sec
Range	2,176 NM	2,167 NM
Altitude	498 NM	499 NM
Velocity (relative to earth)	17,605 ft/sec	17,600 ft/sec
<u>RE-ENTRY</u>		
Time	1,720 sec	1,721 sec
Altitude	300,000 ft	300,000 ft
Range	4,269 NM	4,270 NM
Velocity (relative to earth)	21,458 ft/sec	21,459 ft/sec
Flight path angle (relative)	-19.59°	-19.60°
<u>IMPACT</u>		
Time	1,760 sec	1,761 sec
Longitude	14.4 deg W	14.4 deg W
Latitude (geodetic)	8.3 deg S	8.3 deg S
Range	4,400 NM	4,400 NM
<p>a. Series E R&amp;D missile carrying a MK 4 RV (3829#), AFSWC pod (100#), ADF Mod I decoy pod (190#).</p> <p>b. Series E R&amp;D missile carrying a MK 4 RV (3829#), AFSWC pod (100#), and SP pod (500#)</p> <p>c. Series E R&amp;D missile carrying a MK 4 RV (3829#), AFSWC pod (100#), ADF Mod I decoy pod (190#), and</p> <p>d. Series E R&amp;D missile carrying a MK 4 RV (3829#), ADF Mod I decoy pod (190#), and dual SP pods (100#)</p> <p>e. Series E R&amp;D missile carrying a MK 4 RV (3829#), ADF Mod II decoy pod (375#), and dual SP pods (100#)</p> <p>f. Series E R&amp;D missile carrying a MK 4 RV (3829#), ADF Mod I decoy pod (190#), and dual SP pods (100#)</p> <p>g. Series E R&amp;D missile carrying a MK 5 RV (ballasted to 2829#) and dual SP pods (1000#) or carrying a M</p>		

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FLIGHT PARAMETERS (Continued)

VALUE					
b	c	d	e	f	g
174.1 NM 17.5° 0.08 g's 433.8 NM	173.8 NM 17.5° 0.08 g's 435.1 NM	173.8 NM 17.5° 0.08 g's 434.9 NM	173.5 NM 17.5° 0.08 g's 434.8 NM	196.7 NM 22.7° 0.08 g's 422.7 NM	177 NM 16.4° 0.10 g's 472.1 NM
961 sec .167 NM 499 NM .600 ft/sec	961 sec 2,177 NM 497 NM 17,617 ft/sec	961 sec 2,177 NM 497 NM 17,615 ft/sec	961 sec 2,177 NM 496 NM 17,621 ft/sec	1,047 sec 2,197 NM 655 NM 16,090 ft/sec	1,279 sec 3,125 NM 677 NM 18,371 ft/sec
.721 sec 000 ft 270 NM .459 ft/sec 19.60°	1,720 sec 300,000 ft 4,270 NM 21,461 ft/sec -19.55°	1,720 sec 300,000 ft 4,270 NM 21,461 ft/sec -19.55°	1,719 sec 300,000 ft 4,270 NM 21,461 ft/sec -19.54°	1,886 sec 300,000 ft 4,289 NM 21,316 ft/sec -24.71°	2,354 sec 300,000 ft 6,185 NM 23,214 ft/sec -17.38°
761 sec 14.4 deg W 1.3 deg S 1,400 NM	1,760 sec 14.4 deg W 8.3 deg S 4,400 NM	1,760 sec 14.4 deg W 8.3 deg S 4,400 NM	1,760 sec 14.4 deg W 8.3 deg S 4,400 NM	1,919 sec 14.4 deg W 8.3 deg S 4,400 NM	2,397 sec 13.2 deg E 28 deg S 6,350 NM

190#), and SP pod (500#)

0#), and dual SP pods (1000#).

ods (1000#)

ods (1000#)

ods (1000#) (lofted trajectory)

ring a MK 5 RV (ballasted to 3229#), AFSWC pod (100#), and SP pod (500#).

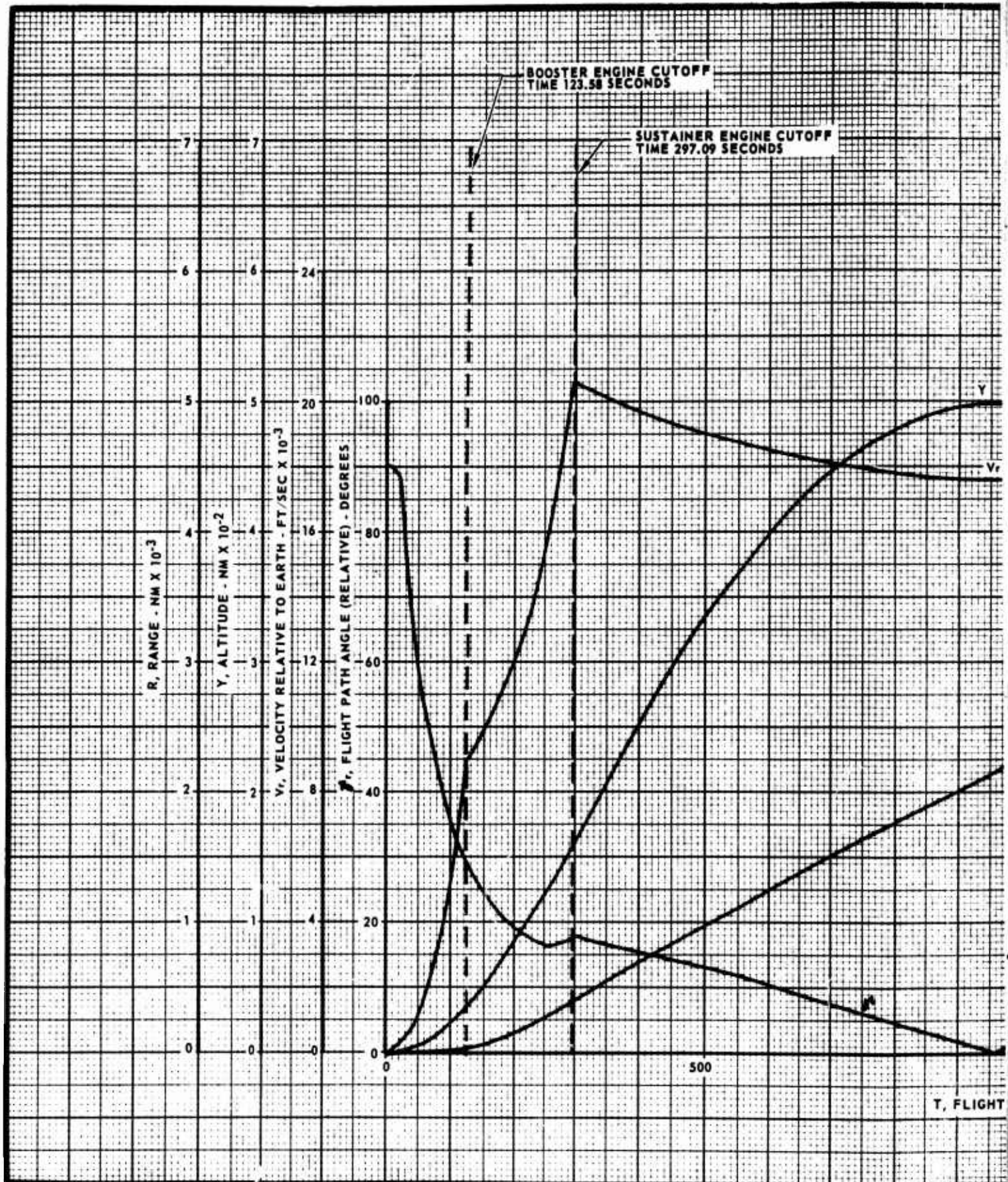
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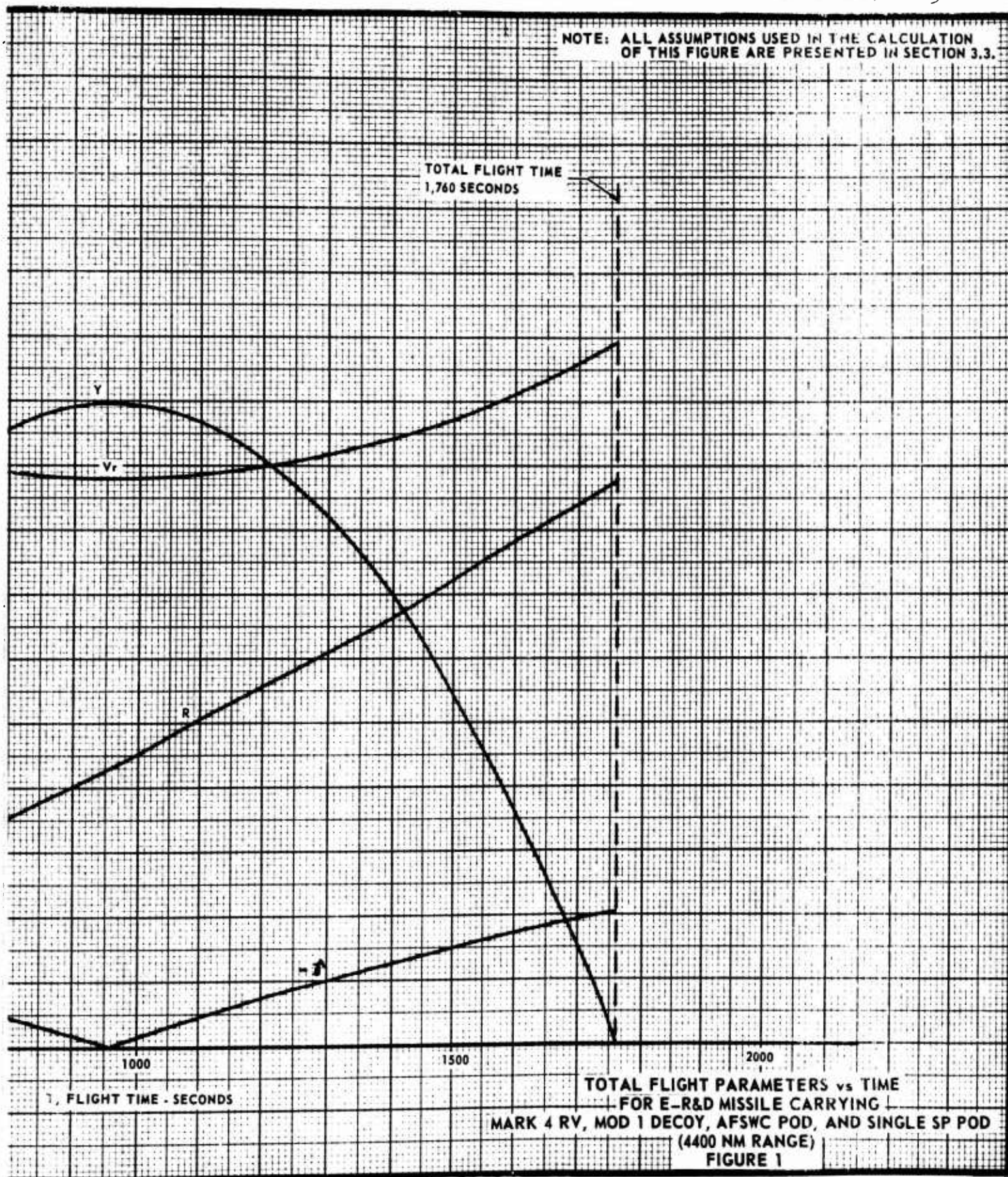
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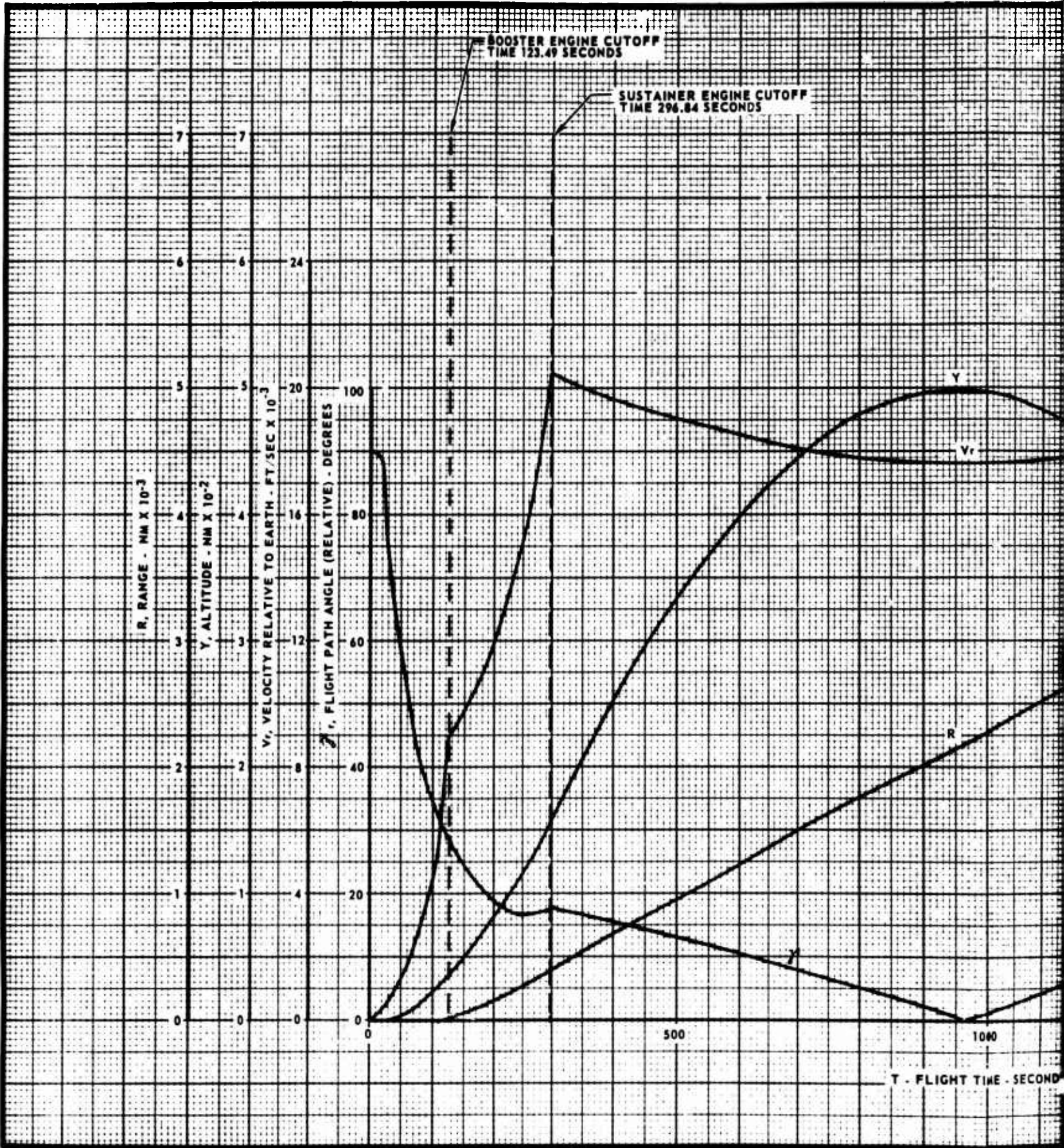
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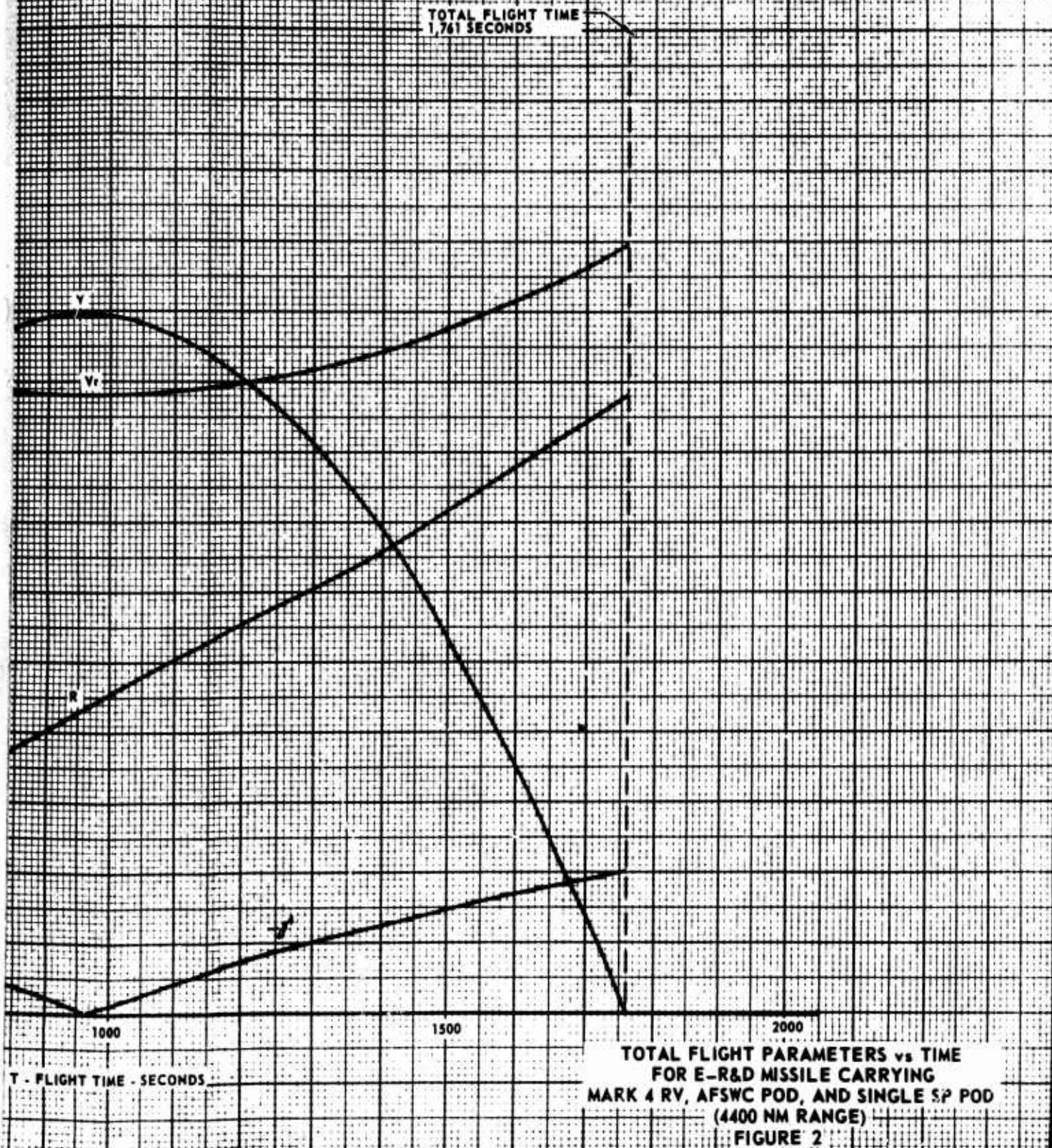
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TOTAL FLIGHT TIME  
1,761 SECONDS



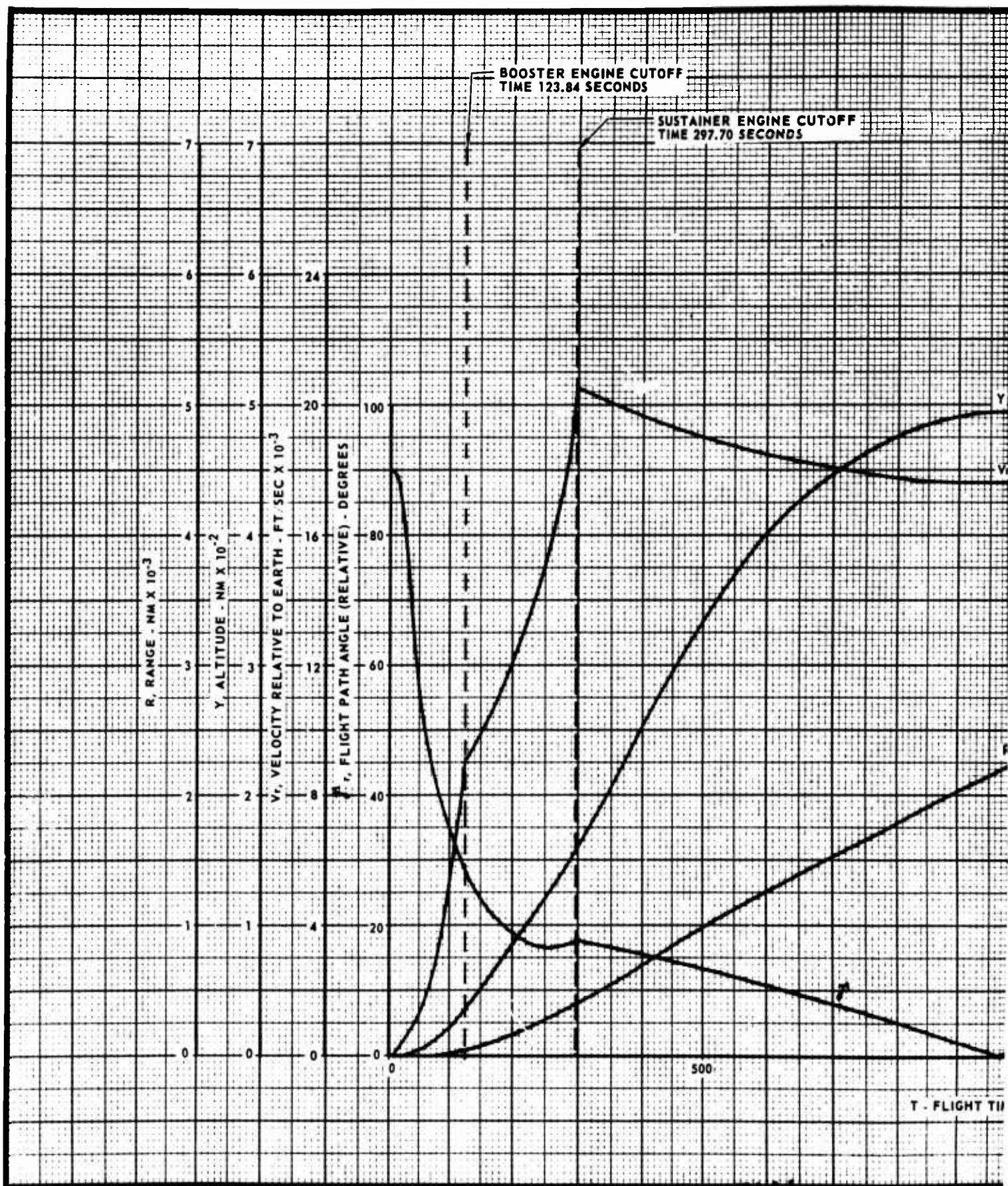
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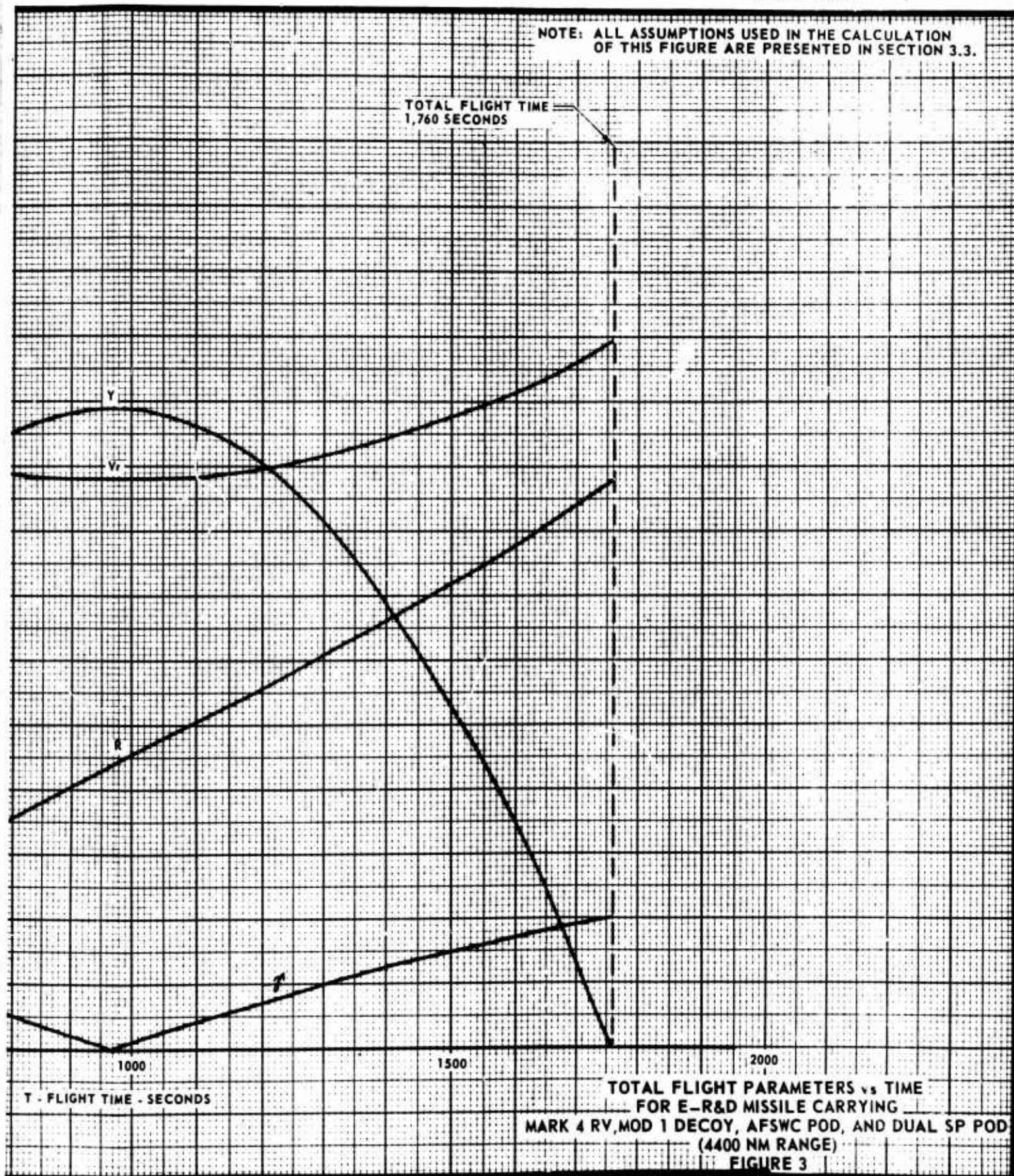


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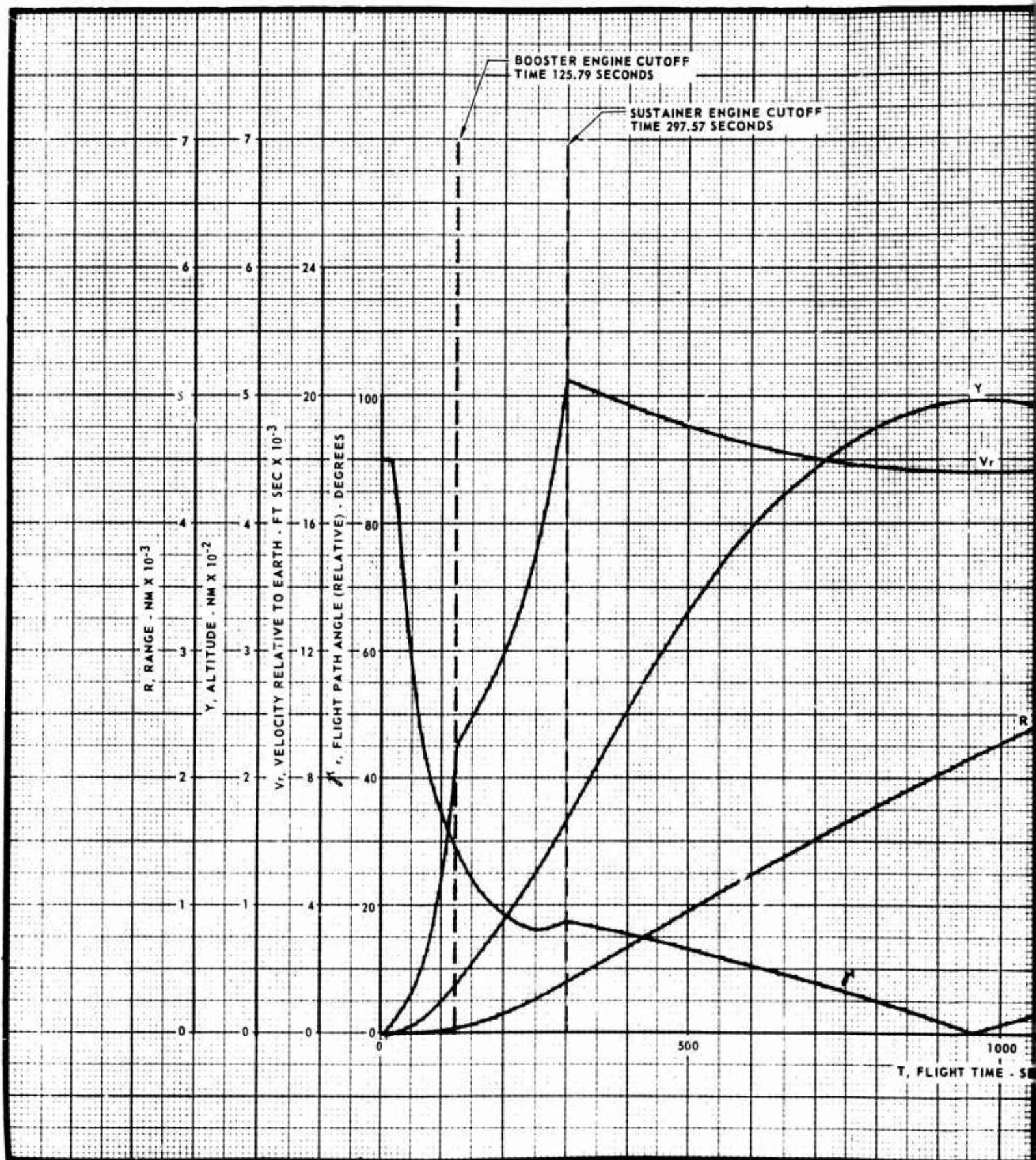
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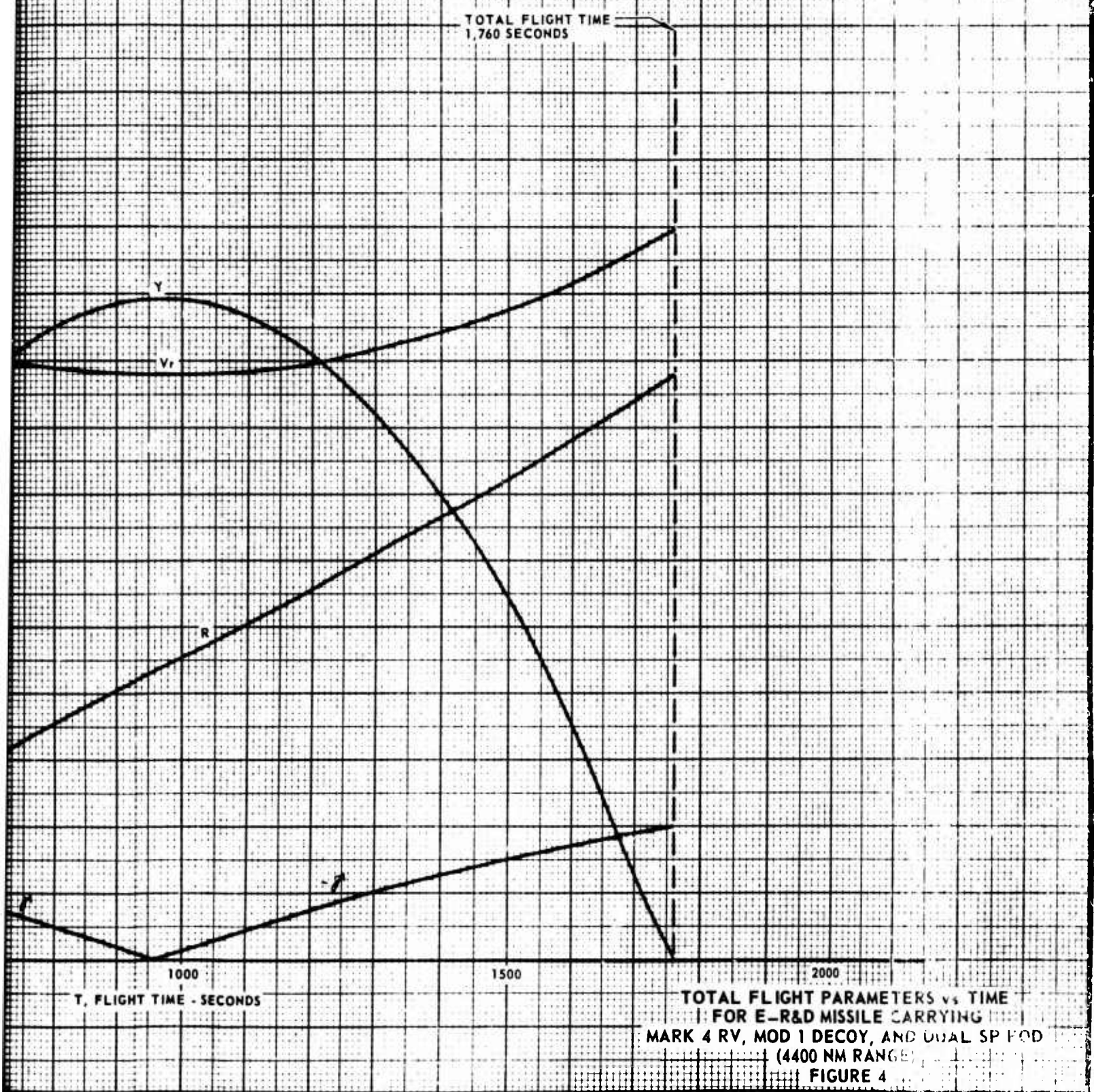


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TOTAL FLIGHT PARAMETERS vs TIME  
FOR E-R&D MISSILE CARRYING  
MARK 4 RV, MOD 1 DECOY, AND DUAL SP POD  
(4400 NM RANGE)  
FIGURE 4

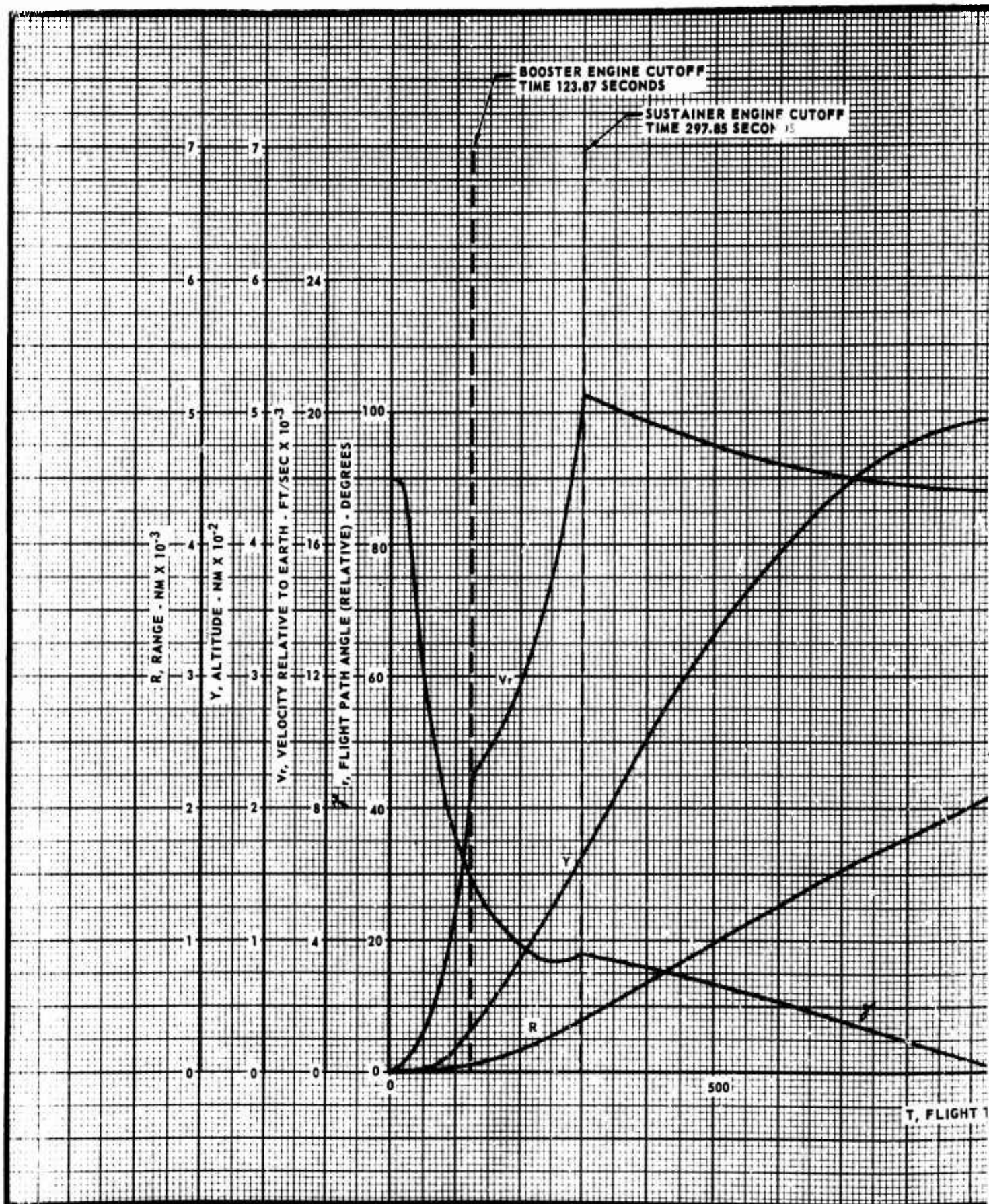
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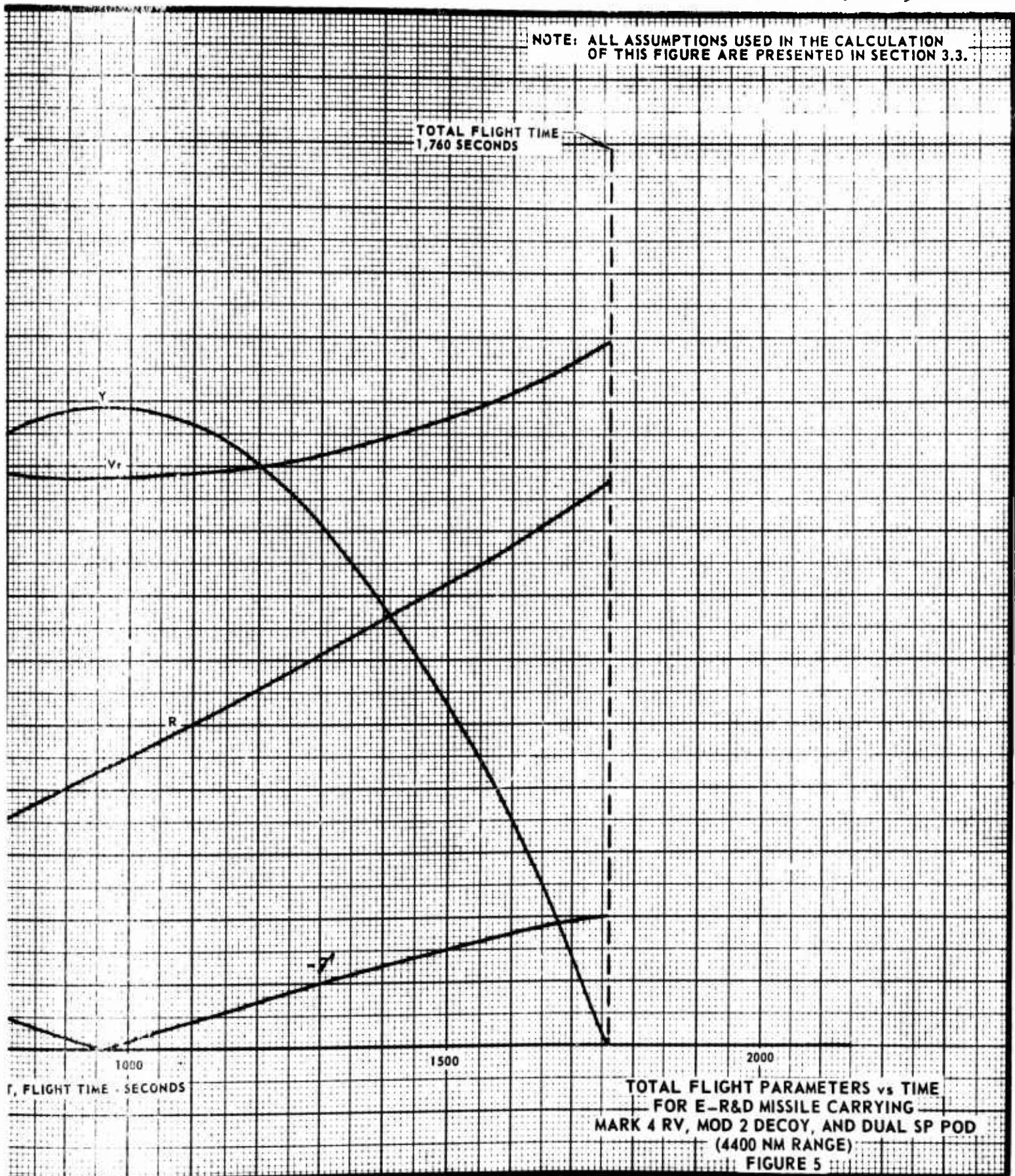


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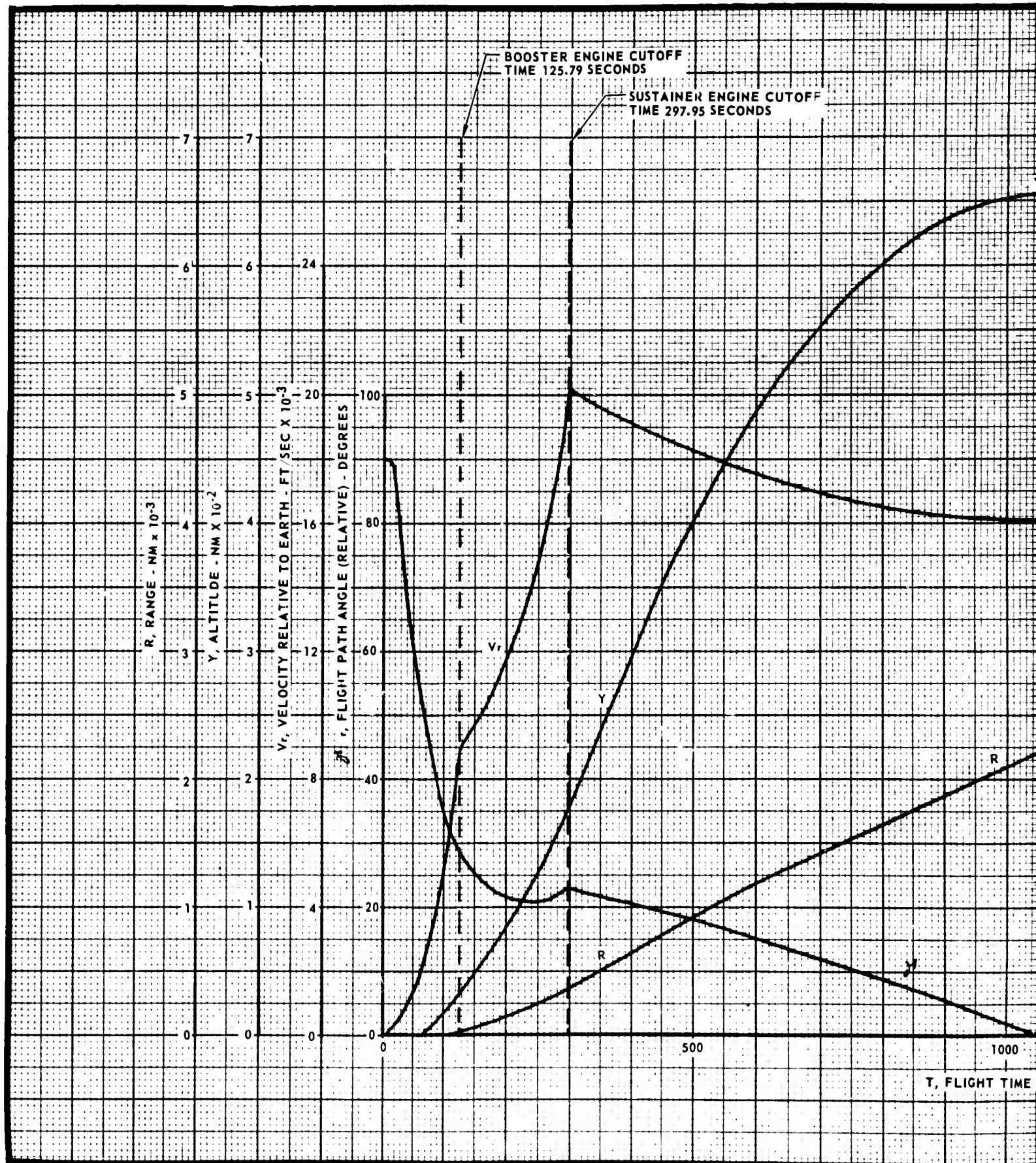
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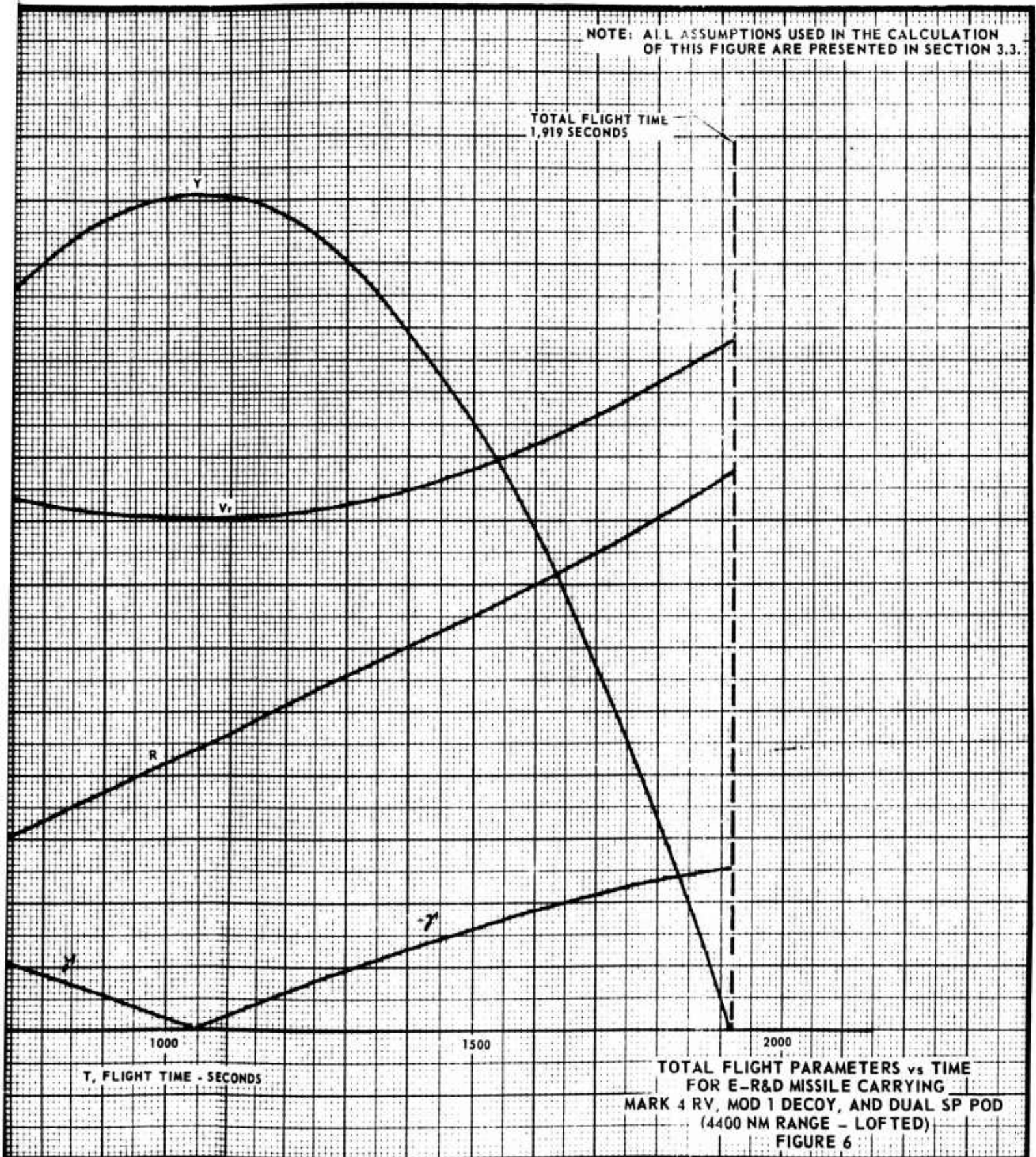
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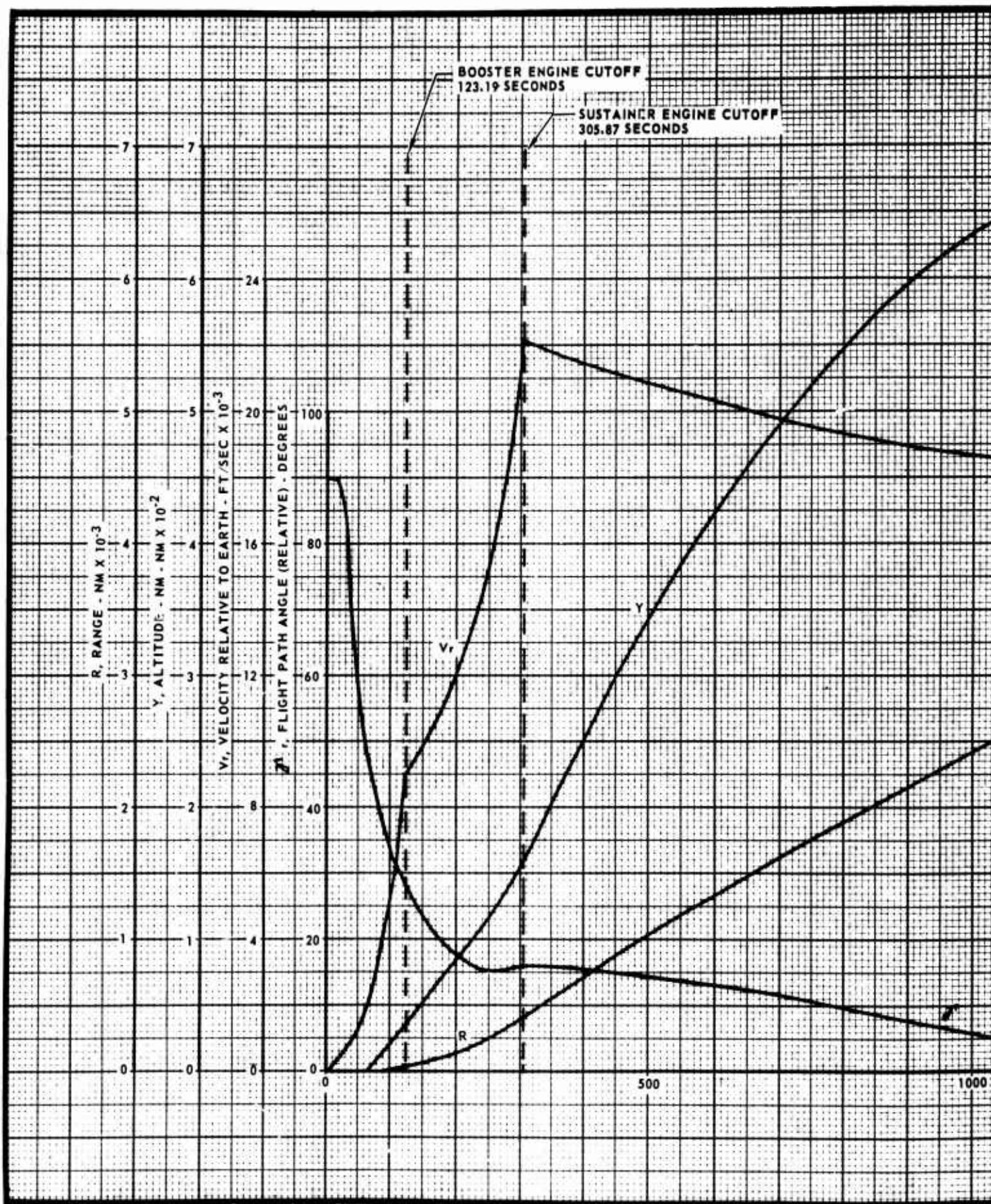


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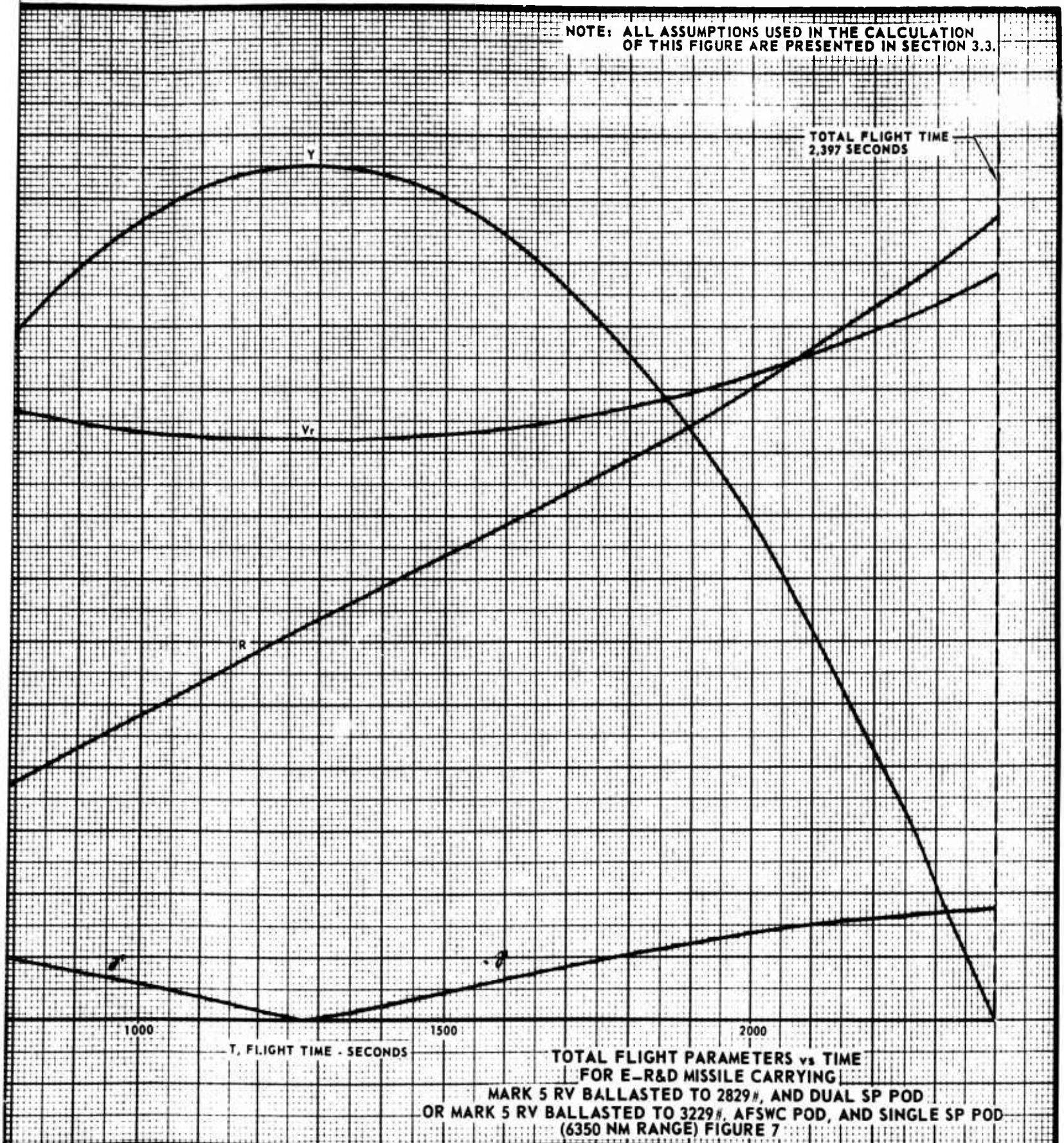


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NOTE: ALL ASSUMPTIONS USED IN THE CALCULATION  
OF THIS FIGURE ARE PRESENTED IN SECTION 3.3.



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